Amendments To The Claims:

Please amend the claims as shown.

1 – 11 (canceled)

12. (new) A gas turbine engine, comprising:

a rotationally mounted rotor having a longitudinal axis;

an axial compressor arranged coaxially along the rotor that produces a compressed intake fluid flow;

a combustion chamber arranged downstream of the compressor which receives the fluid flow and a fuel, and combusts the fluid flow and the fuel to form a hot working medium flow;

a turbine that receives and extracts mechanical energy from the hot working medium flow;

a rotationally fixed inner casing wherein the hot working medium flows through a passage within the inner casing, the inner casing comprising:

a front ring having a collar portion extending in the axial direction arranged coaxially with the rotor, and

a rear ring having a collar portion extending in the axial direction arranged coaxially and down stream of the first ring with respect to the direction of flow of the working medium, the two rings forming an annular gap in an area where the collars partially overlap, and

a spring element arranged to seal the annular gap from the hot working medium having a first end, a second end and a spring region arranged between the first and second ends, the first end secured in a circumferential grove of either the front ring or the rear ring and the second end in intimate contact with a bearing surface of the collar of the other inner casing ring sealing the annular gap from the hot working medium.

13. (new) The engine as claimed in claim 12, wherein the inner casing diverges conically toward the rotor in the direction of flow.

14. (new) The engine as claimed in claim 12, wherein the front ring has a radially inner collar and the rear ring has a radially outer collar.

15. (new) The engine as claimed in claim 14, wherein the front ring forms the radially outer collar and the rear ring forms the radially inner collar such that the annular gap extends in the direction of flow of the working fluid.

16. (new) The engine as claimed in claim 12, wherein the radial width of the circumferential groove is less than twice the material thickness of the spring seal.

17. (new) The engine as claimed in claim 16, wherein the first end of the spring element is connected to the circumferential groove by welding or soldering.

18. (new) The engine as claimed in claim 12, wherein an annular bearing surface is provided on the radially inner collar on a side opposite the working medium.

19. (new) The engine as claimed in claim 12, wherein the spring seal element has a S-shaped cross section.

20. (new) The engine as claimed in claim 12, wherein a cooling medium exerts a higher pressure on an outer diameter surface of the spring seal element relative to the pressure exerted on the inner diameter side by the hot working medium.

21. (new) A gas turbine hot gas sealing system comprising:

a first component having a collar portion;

a second component having a collar portion adjacent the first component collar portion, the first and second collar portions partially overlapping to form an annular gap, the second component having a circumferential grove open to the annular gap; and

an annular spring seal element arranged to seal the annular gap from a hot gas in the turbine having a first end, a second end and a spring region arranged between the first and second ends, the first end region secured within the circumferential grove and the second end in

direct contact with the collar of the first inner casing ring sealing the annular gap from the hot gas.

22. (new) The sealing system as claimed in claim 21, wherein the circumferential grove is facing the annular gap.

23. (new) The sealing system as claimed in claim 21, wherein the first end of the spring element is welded or soldered to the to the circumferential grove.

24. (new) The sealing system as claimed in claim 21, wherein the spring element has an S-shaped cross section.

25. (new) A compliant turbine hot gas seal system, comprising:

a first hot gas component having an annular surface concentric with a centerline of the turbine;

a second hot gas component having a recessed surface arranged proximal and concentric to the annular surface of the first component defining a hot gas gap between the first and second components; and

a seal component having a first and a second portion, the first portion arranged within the recess and the second portion in sliding contact with the annular surface wherein the seal component is pre-stressed in the radial direction and exerts a contact pressure against the annular surface to prevent a flow of hot gas through the hot gas gap.

26. (new) The seal system as claimed in claim 25, wherein the first portion is welded or soldered to the second component.

27. (new) The seal system as claimed in claim 25, wherein the seal element is S-shaped.

28. (new) The seal system as claimed in claim 25, wherein the seal accommodates radial and axial relative motion between the first and second components.

Amendments To The Abstract:

In the English translation document, please add the section heading at page 15 line 1, as follows:

--ABSTRACT

A gas turbine, with a fixed inner housing, arranged concentric to the rotor, with a through flow of working medium, is disclosed. The housing comprises at least two serial rings with an annular gap left between two directly adjacent rings, whereby an annular sealing means is arranged in at least one peripheral groove for sealing the annual gap. According to the invention, a sealing means is provided which permits a greater movement of both components forming the gap, whereby the annual gap is formed by partly overlapping rings, running against the flow direction of the working fluid in the radial sense and the front most of the two rings, in the sense of the flow direction, comprises a locating annular surface for the sealing means embodied as an annular spring element on which the spring element rests under tension such as to seal the annular gap.--